THE RUSSIAN FEDERATION MINISTRY OF TRANSPORT OF THE RUSSIAN FEDERATION FEDERAL AIR TRANSPORT AGENCY

AIRWORTHINESS DIRECTIVE (AD)

March 1, 2024

No.2024-FATA-02074(16-Д)-01

Applicability - PS-90A, PS-90A-76, PS-90A1 Type Engines

Design Organization Country – the Russian Federation

The corrective actions described in the present AD are mandatory. No operator may operate an aircraft subject to this AD other than in accordance with the requirements of this AD.

Subsequent to the flight accident that occurred on 07.12.2023 with the Tu-204C aircraft (RA-64024) associated with the in-flight shutdown of the PS-90A engine No.3949042232031B1 to support continued airworthiness of the PS-90A, PS-90A-76, PS-90A1 engines in order to ensure flight safety of the Russian Federation civil aviation aircraft fleet and based on the Technical Decision of UEC-Aviadvigatel No.PIII-0018-2024 dd 30.01.2024 (hereinafter – the Technical Decision)

IT IS HEREBY SUGGESTED THAT:

- 1. The operators of the PS-90A, PS-90A-76, PS-90A1 engines should perform an unscheduled inspection of all stages of the high pressure compressor of the engines specified in the Technical Decision and the accumulated operating time of which is greater than 4000 flight hours during the nearest scheduled periodic aircraft maintenance check.
- 2. The unscheduled inspection should be performed in accordance with the recommendations contained in the Technical Decision. The results of the works completed should be recorded in the inspection logs to be distributed to the engines design and manufacturing organizations.
- 3. The Airworthiness Directive (AD) should enter into force on 12.03.2024 and terminate upon fulfillment of the provisions of paragraphs 1-2 hereof.

Attachment: Technical Decision No.PШ-0018-2024 dd 30.01.2024, 31 pages.

A.A. Dobryakov

Deputy Director of Federal Air Transport Agency

АКЦИОНЕРНОЕ ОБЩЕСТВО «ОДК-АВИАДВИГАТЕЛЬ»

CONCURRED:

APPROVED:

Начальник 209 ВП МО РФ

Заместитель генерального конструктора - главный конструктор авиационных двигателей

А.В. Бусыгин

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Главный инженер

АО «ОДК-ПМ»

В.Т. Хайрулин

В.Б. Мишенин

DECISION

on Continued Airworthiness of the PS-90A, PS-90A-76, PS-90A1 Engines

January 30, 2024

No. РШ-0018-2024

Perm

3 pages

The flight accident with the PS-90A engine No.3949042232031B1 on the Tu-204C aircraft (RA-64024) occurred on 07.12.2023.

According to the Maintenance Manual, the routine inspection operations at all HPC stages are to be performed at the following engines:

- PS-90A and PS-90A1 after 3 600 flight hours (then every 1 000 flights hours);
- PS-90A-76 after 4 000 flights hours (then every 1 000 flights hours).

DECIDED:

An unscheduled inspection of all HPC stages is to be performed on the PS-90A, PS-90A-76, PS-90A1 engines in accordance with the list below (the operating time of which is greater than ~4 000 flight hours and the nearest scheduled maintenance check of which doesn't envisage any regular HPC inspections) on the following engine fleet in Russian airlines:

Operator	Engine types	Engine number	Operating tir Engine number TTSN /TSO		
			TTSN	TSO	
"Rossiya" Government		3949044622072 Л1	4741	-	
Squadron (Federal State Budgetary	PS-90A	3949042002011 Л2Р5	22557	3973	
Institution)		3949044002031 B1P3	11537	5332	
Russian Post		3949041032009 В1Р1Л1	14291	4393	
(Federal State Unitary	PS-90A	3949041922001 P1	14674	6972	
Enterprise)		3949042922004 P1	16059	9240	
		3949854102049 P2	15307	4954	
		3949853922060 P1	10937	5562	
Volgo Dagos Aislings	DC 004 76	3949854602007 Л1Р3	22959	6656	
Volga-Dnepr Airlines	PS-90A -76	3949853002024 B1P3	16600	7794	
		3949851102010 B1P2	14783	8292	
		3949852302019 В1Л1Р4	25923	10640	
Aviactor TII Airlines	DC OOA	3949043202029 Л1Р2В2	19988	6159	
Aviastar-TU Airlines	PS-90A	3949043201031 Л1Р7В3	36986	6394	

and oversees airlines:

Operator	Engine types	Engine number	Operating TTSN /TS	-
			11311	130
		3949041722077Л1Р1	12602	5321
		3949042622055 P2	15376	5440
Cubana de Aviacion S.A.	PS-90A	3949044101057 P9	43839	6786
Cubana de Aviación S.A.	F3-30A	3949041102003 В1Л2Р6	43031	7314
		3949043202027 P2	10769	7500
		3949042522030 P1	14976	8250

The inspection may be combined with the nearest scheduled maintenance check.

The inspection is to be performed in accordance with the Engine Operation Manual:

PS-90A: 94-00-807P3 Section 072.30.00 Page 233 Task Card "Visual & Optical Inspection of PS-90A HPC Blades and Spacer Rings";

PS-90A-76: 85-00-800P9 Section 072.30.00 Page 245 Task Card "Visual & Optical Inspection of PS-90A HPC Blades and Spacer Rings".

to Maintenance Manual	TASK CARD	Pages 245252e		
Maintenance Manual Item 072.30.00B	Task Title VISUAL & OPTICAL INSPECTION OF HPC BLADES AND SPACER RINGS (FIGURES 206, 207)	Manpower6.0MH.		
	Description of Operations and Technical Requirements	Works to be performed in case of noncompliance with Technical Requirements	Check	
2 GENERAL 3 PERFORM SHUTDO 4 IT IS PRO THE ROT 5 ONLY EM AND WH INSPECT 6 PERFORM PERSONS CONTRO 7 FLAWS S VISUALL 8 IT IS PRO	IEASURES – 072.00.00, PAGE 203. REQUIREMENTS TO INSTALLATION WORKS – 072.00.00, PAGE 413. I THE INSPECTION AT LEAST IN TWO HOURS AFTER ENGINE WN HIBITED TO INSERT, RETRACT AND ADJUST THE ENDOSCOPE DURING ATION OF THE HPC ROTOR. IPLOYEES WHO HAVE A PERMIT FOR OPTICAL INSPECTION WORKS O HAVE STUDIED SAFETY MEASURES ARE ALLOWED TO PERFORM SON WORKS. If THE WORKS IN THE FAN DUCT WITH A GROUP OF TWO-THREE SONE OF WHICH MUST ALWAYS STAY NEAR THE THRUST REVERSER LEVER AND NOT ALLOW ITS OCCASIONAL MOVEMENT. SIZE AND CHARACTERISTICS ARE DETERMINED BY THE INSPECTOR Y BASED ON THE EXPERIENCE. OHIBITED TO ROTATE THE HPC ROTOR IF THE FUEL FILTERS ARE DEFROM THE ENGINE.			
	INSTALL A BANNER PROHIBITING ENGINE STARTING AND CONNECTION OF THE GROUND HYDRAULIC PLANT TO THE ENGINE HYDRAULIC SYSTEM IN THE COCKPIT. 2 SECURE THE THRUST REVERSER IN THE "FORWARD THRUST" POSITION (072.80.00, PAGE 209).			

		Description of Operations and Technical Requ	irements		Works to be performed in case of noncompliance with Technical Requirements	Chec
•	are a set of replaceable parts	s in accordance with Table 1.				
Pos.	Designation	Name	Q-ty	Note		
	H08-440-02-6	Locking washer	4	New		
	94-03-1392	Gasket	2	New		
		tion using an optical instrument tec				
Prep Note	oare an endoscope for operation e — In the process of operation cloth wetter	on wipe the surfaces of the optical with alcohol and wrung out.	Page 401) and prepare	re a portable light.		
Prep Note	oare an endoscope for operation	on wipe the surfaces of the optical with alcohol and wrung out.	Page 401) and prepare	re a portable light.		
Note Insp	oare an endoscope for operation e — In the process of operation cloth wetter	on wipe the surfaces of the optical with alcohol and wrung out. m the leading edge side:	Page 401) and prepare	re a portable light.	Repair the locks	
Prep Note Insp .1 Re	pare an endoscope for operation of the process of operation cloth wetter each HPC 1st stage blades from the emove panels "1-4" and "1-5".	on wipe the surfaces of the optical with alcohol and wrung out. m the leading edge side:	Page 401) and prepare thinical description. elements of the endo	re a portable light. oscope with a clean	Repair the locks (072.70.00, Page 203)	

Notes - 1 The IGV plug is marked "HPC" on the end surface.

2 During the inspection, turn the HPC rotor with an electric drive or a handle for turning the HPC rotor. Determine the completion of one revolution of the HPC rotor by the pulse counter on the remote control of the electric drive or visually mark the first blade to be checked. 3 The number of blades in the 1st stage impeller – 34 pcs. 5.4 Insert the endoscope in the hatch 1 between IGV vanes with the direction of view towards the nozzle. Inspect the HPC 1st stage blades from the leading edge side, vibration shrouds and vibration shroud joints. CAUTION. TAKE PRECAUTIONS DURING BLADES INSPECTION, ELIMINATING THE FAILURE OF THE ENDOSCOPE, INGESTION OF ITS PARTS AND FOREIGN OBJECTS IN THE ENGINE FLOWPATH Blades damage acceptable for operation without rectification. a) The presence of a smooth bend of corner of the 1st stage blade profile from the leading edge side (Figure 206, Table 2): the total number of blades with bend in the impeller – ten, maximum; adjacent – three, maximum. b) The presence of nicks with the depth of 0.1 mm, maximum: on the airfoil profile section (pressure side, suction side) and blade leading and trailing edges; on vibration shrouds (for 1st and 2nd stage blades). c) The presence of leading and trailing edge damage – two nicks on each. In total on both edges – three nicks, maximum.		Description of Operations and Technical Requirements	Works to be performed in case of noncompliance with Technical Requirements	Chec
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 on the airfoil profile section (pressure side, suction side) and blade leading and trailing edges; on vibration shrouds (for 1st and 2nd stage blades). c) The presence of leading and trailing edge damage – two nicks on each. 	- adjacent -	- three, maximum.		
	- on the air	foil profile section (pressure side, suction side) and blade leading and trailing edges;		

Description of Operations and Technical Requirements	Works to be performed in case of noncompliance with Technical Requirements	Check
CAUTION. NOT ACCEPTABLE:		
1 SHARP EXTENDING EDGES OF NICKS. RECTIFY IN ACCORDANCE WITH INSTRUCTION 94-01-805И250.		
2 CRACKS IN THE LOCATIONS OF BENDS AND NICKS ON THE BLADE AIRFOIL SECTION AND EDGES. CRACKS PRESENCE CHECK - VISUAL.		
3 NICKS WITH BULGING METAL ON THE BACK SIDE OF THE BLADE.		
4 NICKS IN ONE CROSS SECTION OF THE AIRFOIL AT BLADE LEADING AND TRAILING EDGE.		
5 NICKS IN THE SECTIONS WITH THE DISTANCE BETWEEN THEM OF LESS THAN 10 MM.		
6 NICKS ON THE AIRFOIL TIP.		
7 TORN NICKS AND BENDS OF THE BLADE AIRFOIL SECTION .		
8 UNSHROUDING.	Eliminate HPC 1 st stage	
	blade unshrouding (072.30.00, Page 469)	
5.5 Mark the blades with the damage acceptable for operation after rectification in accordance with Instruction 94-01-805И250.	(072.30.00, Fage 409)	
CAUTION. PRESENT THE BLADES DAMAGE BEYOND RECTIFICATION LIMITS TO THE ENGINE MANUFACTURER REPRESENTATIVE TO TAKE DECISION ON ENGINE FURTHER OPERATION OR REPAIR.		
5.6 Carefully retract the endoscope from the hatch.		
5.7 Put the plug in the inspection hatch 1. Press and sink it and turn it clockwise by 90°. Pull the plug all the way out. Make sure that the marks on the plug and the hatch body are aligned. Unscrew and remove the wrench from the plug.		
CAUTION. DEPRESSION OF THE END SURFACE OF THE PLUG RELATIVE TO THE SURFACES "P" AND "C" IS NOT ACCEPTABLE .		

	Description of	Operations and Technical Requirements		Works to be performed in case of noncompliance with Technical Requirements	Chec
6 Inspect the	e HPC blades of the downstream	stages and stage 312 spaces	r rings		
Notes -	1 The plug has a marking of	on the inner end surface corres	ponding to the stage number		
	2 Perform plug removal, successively for each sta	1 1 0	tallation back into its position		
	3 Number of blades in im	pellers: 1 stage – 34 pcs.			
	Stage $2-35$ pcs.	Stage $6-75$ pcs.	Stage 10 – 92 pcs.		
	Stage $3 - 39$ pcs.	Stage $7 - 82$ pcs. Stage $8 - 91$ pcs.	Stage $11 - 101$ pcs.		
	Stage $4-45$ pcs.	Stage 8 – 91 pcs.	Stage 12 – 105 pcs.		
	Stage $5 - 67$ pcs.	Stage $9-95$ pcs.	Stage $13 - 103$ pcs.		
6.1 Remove	panels "2-3", "2-4", "3-3", "3-4"	' (072.70.00, Page 407).			
6.2 Remove	the plug from the inspection hat	ch of the respective stage:			
6.2.1 For insp	pection hatches at the 1st stage val	ne assembly 2 and 3 rd stage va	nne assembly 3 – similar to Item 5.3		
Notes	 If hatches with threaded plug proceed as follows: 	gs are used on the case with H	PC 1 st , 2 nd ad 3 rd stage vanes,		
	1) Unlock the inspection	n hatch plug.			
	2) Install the wrench 19 of the plug.	9-590 assembled with the adap	oter and torque wrench in the recess		
		ounterclockwise, and carefully			
	pection hatches at the 512 th sta	age vane assemblies proceed a	s follows::		
	ock the inspection hatch plug;				
	11.4 1.20.07.000 4	olug:			
- insta	all the wrench 32-27-899 on the perew the plug, counterclockwise, as				

	Description of Operations and Technical Requirements	Works to be performed in case of noncompliance with Technical Requirements	Ch
and inspec	endoscope in the hatch between the guide vanes with the direction of view towards the air intake et the blades from the trailing edge side. Turn the endoscope 180 degrees and inspect the next es from the leading edge side.		
Notes –	1 Inspect the blades similar as described in Items 5.4, 5.5 as relevant for each stage.		
	2 Acceptable for operation without rectification are the blades with corner bends at the tip part of the blade with the size of 1x1 mm, max:		
	- 1 st stage blades from the trailing edge side;		
	- 213 th stage blades at the leading and trailing edges.		
- Stages 3 - Stages 6	2 – 6 blades, max; 4, 5 – 8 blades, max; 7, 8 – 10 blades, max; 13 – 12 blades, max;		
6.5 Inspect 3	12th stages spacer rings.		
	MARKS OF VANES RUBBING AGAINST THE SPACER RINGS SURFACE AND A		
CAUTION.	VISIBLE GAP BETWEEN THE SPACER RING AND THE IMPELLER DISK ARE NOT ACCEPTABLE.		
CAUTION. 6.6 Carefully	VISIBLE GAP BETWEEN THE SPACER RING AND THE IMPELLER DISK ARE NOT		

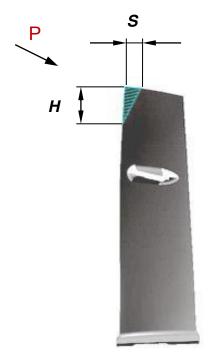
Description of Operations and Technical Requirements	Works to be performed in case of noncompliance with Technical Requirements	Chec
6.7.1 For inspection hatches at the 1 st stage vane assembly 2 and 3 rd stage vane assembly 3 – similar to Item 5.7.		
Note – If hatches with threaded plugs are used on the case with HPC 1 st , 2 nd ad 3 rd stage vanes, proceed as follows:		
1) Install the plug on the inspection hatch of the respective stage and screw it clockwise by hand.		
2) Tighten the plug with the wrench 19-590, the adapter and the torque wrench. The torque is 2.02.5 kgf·m.		
3) Lock the plug with wire – Type 44 (070.00.03, Page 1, Figure 18).		
6.7.2 For inspection hatches at the 512th stage vane assemblies proceed as follows:		
a) install the plug of the respective stage on the inspection hatch and screw it clockwise by hand;		
b) tighten the plug with the wrench 32-27-899, the adapter and the torque wrench. The torque is 2.02.5 kgf·m.		
c) lock with wire the following plugs:		
- inspection hatch plugs 4, 5, 6 – Type 44 (070.00.03, page 1, Figure 18);		
- inspection hatch plugs 7, 10 – Type 41 (070.00.03, page 1, Figure 15);		
- inspection hatch plugs 8, 9 - Type 42 (070.00.03, page 1, Figure 16).		
7 Inspect the 13th stage blades from the trailing edge side:		
7.1 Unlock, unscrew and remove the bolts 12 (2 pcs.) used to secure the plug 11 of combustor diffuser hatch from the engine left or right side. Remove the plug with a gasket.		
7.2 Remove the gasket from the plug.		
7.3 Insert the endoscope in the hatch of combustor diffuser and install between the nozzle vanes with the direction of view towards the air intake. Inspect the 13 th stage blades from the trailing edge side		

Description of Operations and Technical Requirements	Works to be performed in case of noncompliance with Technical Requirements	Chec
7.4 Carefully retract the endoscope from the hatch		
8 Install the plug 11 on the combustor diffuser hatch::		
8.1 Install a new gasket in the plug 11 recess		
8.2 Install the plug 11 on the combustor diffuser hatch and secure it with bolts 12 (2 pcs.) with new locking washers.		
8.3 Tighten the bolts 12 used to secure the plug 11. The torque is 0.81.3 kgf·m.		
8.4 Lock the bolts with new locking washers (070.00.03, Page 1).		
9 Install the removed panels "1-4", "1-5", "2-3", "2-4", "3-3" and "3-4" (072.70.00, Page 411).		
10 Remove the fixture for works in the fan duct 94–27–893 (072.70.00, Page 401).		
11 11Remove the handle or electric drive for HPC rotor turning from the engine accessory drive gearbox (72.00.00, Page 611 or 613).		
ВНИМАНИЕ. 1 UNLOCK THE THRUST REVERSER IN THE "FORWARD THRUST" POSITION (072.80.00, PAGE 209).		
2 TAKE AWAY THE BANNER PROHIBITING ENGINE STARTING AND CONNECTION OF THE GROUND HYDRAULIC PLANT TO THE ENGINE HYDRAULIC SYSTEM FROM THE COCKPIT.		
12 Make a record of the works performed in the engine log book.		
13 Generate an HPC blades inspection report (see Annex A) in three copies, forward one copy to both the engine design and engine manufacturing organizations.		

Description of Operations and Technical Requirements			Works to be performed in case of noncompliance with Technical Requirements	
Instrumentation	Tools and fixtures		Consumables	
"OLYMPUS" IPLEX FX video endoscope including: - system unit IV8000-2, with the changeable working part IV8420 (diameter 4 mm, length 2000 mm) with adapters IV84-AT120D/FF and IV84-AT120S/FF - changeable working part IV8620 (diameter 6 mm, length 2000 mm) with adapters IV86-AT120D/FF and IV86-AT120S/FF It is allowed to use endoscopes that meet the technical requirements (see 070.00.07)	Socket wrench head 19-916 S=10 (with internal square 10) Head for HPC plugs removal and installation S=10 32-27-899 Box wrench 19-140 S=10x12 Torque wrench M=0,31,5 kgf·m 84-27-827-01 Torque wrench M=1,05,0 kgf·m 84-27-828 Wrench for quick-release plugs removal 19-601(M8) S_{sq} =10 (inner) Cutting pliers Type 1X2H8 TV39-26-003578 87514-2009 Hammer 19-801 Adapter 93-19-810 Flat-nose pliers 7814-0266 μ H6.X.6 GOST 5547-93 IGV vanes turning device 19-593 -2 pcs. Fixture for works in the fan duct 94-27-893 Swivel handle 19-8028 Portable light 93-27-844 Rod 19-011(extension to the swivel handle 19-8028) Rod 19-8027 S_{sq} =10 (inner) Caulking tool 84-19-022 (for plate lock locking) Wrench for plugs removal 19-590 S=8	GOST 29298-20 Nefras – C50/17 Nefras – C2-80/ TU 38.401-67-1	nade of cotton fabric 005 70 GOST 8505-80 or /120 108-92 ГІМ-221 (ЦИАТИМ-221) 12X18H10T	- 2 pcs 0.5 l - 0.1 kg - 1.5 m - 0.05 l

Table 2 – Values of HPC 1st stage blade airfoil section corner bend at the leading edge side

Blade airfoil section corner bend value			
Height from the tip, (H), mm	Width from the leading edge (S), mm	Bend value, (Δ) mm	
25, max	8, max	1,5, max	



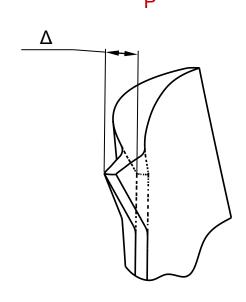
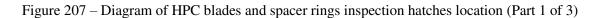
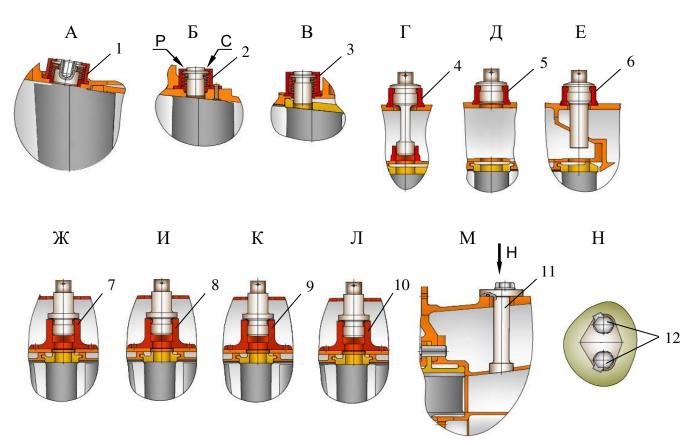


Figure 206 – Diagram of acceptable values of HPC 1st stage blade airfoil section corner bend at the leading edge side





- 1 Stage 1 blade inspection hatch, IGV case
- 2 Stage 1 & 2 blade inspection hatch St.1 vane assembly
- 3 Stage 3 & 4 blade inspection hatch St.3 vane assembly
- 4 Stage 5 & 6 blade inspection hatch St.5 vane assembly
- 5 Stage 6 & 7 blade inspection hatch St.6 vane assembly
- 6 Stage 7 & 8 blade inspection hatch St.7 vane assembly

- $7-Stage\ 9\ \&\ 10\ blade$ inspection hatch St.9 vane assembly
- 8 Stage 10& 11 blade inspection hatch St.10 vane assembly
- 9 Stage 11 & 12 blade inspection hatch St.11 vane assembly
- 10 Stage 12 & 13 blade inspection hatch St.12 vane assembly
- 11 Combustor diffuser hatch plug
- 12-bolt

Figure 207 – Diagram of HPC blades and spacer rings inspection hatches location (Part 2 of 3)

MAINTENANCE MANUAL

Figure 207 – Diagram of HPC blades and spacer rings inspection hatches location (Part 3 of 3)

Works to be performed in case of noncompliance with Technical Requirements

Check

Annex A (mandatory) HPC Blades Inspection Report

Product – PS-90A Engine...

Engine No.: ...

TTSN (hours/ cycles): ...

TSO (hours/ cycles): ...

A/C model: ... Powerplant number: ...

Work type (maintenance, post-flight inspection, etc.): ...

Work completion date (DD. MM. Year): ...

Work performed by (Name): ...

Operator representative (Name): ...

Table 3 – HPC Flowpath Inspection Results

				Damage description *										
Stage mimber	Total number of blades containing damage, pcs.	Total number of blades with rectification, pcs.	Conventional blade No.	Type of damage, (N – nick; D – dent of the airfoil section; B – bend of edge; BC – bend of corner) and	Size from blade tip (l), mm	(K_{max})	ge size., L_{max}),	dent on the	Edge bend size (δ), mm		er beno nax, S _{max}		Concilano	Compnance
			S their	their quantity on the blade		Kmax	Lmax			Hmax	Smax	Δ	Specs	NO

^{* –} Columns should be filled depending on the type of damage (nicks, dents on the airfoil section, edge bending or corner bending), indicating their number and quantity per one blade, and must be accompanied by photos or video records.

to Maintenance Manual	TASK CARD	Pages 233244a			
Maintenance Manual Item	Task Title VISUAL & OPTICAL INSPECTION	Manpower			
072.30.00в	OF HPC BLADES AND SPACER RINGS	6.0 _{MH.}			
	(FIGURES 206, 207)	***			
	Works to be performed in case of noncompliance with Technical Requirements	Check			
	START THE WORKS, BE SURE TO FAMILIARIZE YOURSELF WITH THE SURES, SECTION 072.00.00, PAGE 203.				
WARNING. PERFORM TH	E INSPECTION AT LEAST IN TWO HOURS AFTER ENGINE SHUTDOWN.				
WARNING. 1 DE-ENERGIZ	ZE THE AIRCRAFT ONBOARD ELECTRICITY SUPPLY SYSTEM.				
2 INSTALL A F	BANNER PROHIBITING ENGINE STARTING IN THE COCKPIT.				
3 SECURE THE (073.21.00, PA					
	TED TO INSERT, RETRACT AND ADJUST THE ENDOSCOPE DURING THE FETHE HPC ROTOR.				
2 ONLY EMPL WHO HAVE S INSPECTION					
3 FLAWS SIZE VISUALLY B					
4 IT IS PROHIB REMOVED FI					
CAUTION. GENERAL REQ 413.					
1 Prepare a set of replaceable j	parts in accordance with Table 1.				

Description of Operations and Technical Requirements in case of noncompliance with Technical Requirements Cl
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Table 1 – Kitting of Parts and Components

Pos.	Designation	Name	Q-ty	Note
	H08-440-02-6	Locking washer	4	New
	94-03-1392	Gasket	2	New

- 2 Install a handle or an electric drive for HPC rotor turning (072.00.00, page 607 or 609).
- 3 Install a fixture for the works in the fan duct (072.70.00, page 209), put the mats on the inner surfaces of the front casing and TRU doors, prepare a portable light.
- 4 Prepare an endoscope for operation using an optical instrument technical description.
 - Note In the process of operation wipe the surfaces of the optical elements of the endoscope with a clean cloth wetted with alcohol and wrung out.
- 5 Inspect HPC 1st stage blades from the leading edge side:
- 5.1 Remove panels "1-4" and "1-5" (072.70.00, page 401).
- 5.2 Set the IGV vanes in "0" position by IGV scales by simultaneous rotation of rollers with levers using two fixtures 93-19-811.
- 5.3 Install the wrench 19-601 assembled with an adapter into the threaded hole of the inspection hatch plug 1 on the HPC IGV housing and screw it clockwise. Press the plug, sink it and turn it clockwise by 90°. Remove the plug from the engine.
 - Notes 1 The IGV plug is marked "HPC" on the end surface.
 - 2 During the inspection, turn the HPC rotor with an electric drive or a handle for turning the HPC rotor. Determine the completion of one revolution of the HPC rotor by the pulse counter on the remote control of the electric drive or visually mark the first blade to be checked.
 - 3 The number of blades in the 1^{st} stage impeller 34 pcs.

Repair the locks (072.70.00, page 203)

- 5.4 Insert the endoscope in the hatch 1 between IGV vanes with the direction of view towards the nozzle. Inspect the HPC 1st stage blades from the leading edge side, vibration shrouds and vibration shroud joints.
 - CAUTION. TAKE PRECAUTIONS DURING BLADES INSPECTION, ELIMINATING THE FAILURE OF THE ENDOSCOPE, INGESTION OF ITS PARTS AND FOREIGN OBJECTS IN THE ENGINE FLOWPATH.

Blades damage acceptable for operation without rectification.

- a) The presence of a smooth bend of corner of the 1st stage blade airfoil section from the leading edge side (Figure 206, Table 2):
 - the total number of blades with bend in the impeller ten, maximum;
 - adjacent three, maximum.
- b) The presence of nicks with the depth of 0.1 mm, maximum:
 - on the airfoil profile section (pressure side, suction side) and blade leading and trailing edges;
 - on vibration shrouds (for 1st and 2nd stage blades).
- c) The presence of leading and trailing edge damage two nicks on each. In total on both edges three nicks, maximum.

CAUTION. NOT ACCEPTABLE:

- 1 SHARP EXTENDING EDGES OF NICKS. RECTIFY IN ACCORDANCE WITH INSTRUCTION 94-01-805И250.
- 2 CRACKS IN THE LOCATIONS OF BENDS AND NICKS ON THE BLADE AIRFOIL SECTION AND EDGES. CRACKS PRESENCE CHECK VISUAL.
- 3 NICKS WITH BULGING METAL ON THE BACK SIDE OF THE BLADE.
- 4 NICKS IN ONE CROSS SECTION OF THE AIRFOIL AT BLADE LEADING AND TRAILING EDGE.
- 5 NICKS IN THE SECTIONS WITH THE DISTANCE BETWEEN THEM OF LESS THAN 10 MM.
- 6 NICKS ON THE AIRFOIL TIP.
- 7 TORN NICKS AND BENDS OF THE BLADE AIRFOIL SECTION.
- 8 UNSHROUDING

Eliminate HPC 1st stage blade unshrouding (072.30.00, page 449)

5.5 Mark the blades with the damage acceptable for operation after rectification in accordance with Instruction 94-01-805И250.

Description of Operations and Technical Requirements

CAUTION. PRESENT THE BLADES DAMAGE BEYOND RECTIFICATION LIMITS TO THE ENGINE MANUFACTURER REPRESENTATIVE TO TAKE DECISION ON ENGINE FURTHER OPERATION OR REPAIR.

- 5.6 Carefully retract the endoscope from the hatch.
- 5.7 Put the plug in the inspection hatch 1. Press and sink it and turn it clockwise by 90°. Pull the plug all the way out. Make sure that the marks on the plug and the hatch body are aligned. Unscrew and remove the wrench from the plug.

CAUTION. DEPRESSION OF THE END SURFACE OF THE PLUG RELATIVE TO THE SURFACES "P" AND "C" IS NOT ACCEPTABLE

6 Inspect the HPC blades of the downstream stages and stage 3...12 spacer rings.

1 The plug has a marking on the inner end surface corresponding to the stage number. Notes -

> 2 Perform plug removal, blades inspection and plug installation back into its position successively for each stage.

3 Number of blades in impellers: 1 stage - 34 pcs.

Stage $2-35$ pcs.	Stage 6 – 75 pcs.	Stage 10 – 92 pcs.
Stage $3 - 39$ pcs.	Stage 7 – 82 pcs.	Stage 11 – 101 pcs.
Stage 4 – 45 pcs.	Stage 8 – 91 pcs.	Stage 12 – 105 pcs.
Stage $5 - 67$ pcs.	Stage 9 – 95 pcs.	Stage 13 – 103 pcs.

- 6.1 Remove panels "2-3", "2-4", "3-3", "3-4" (072.70.00, page 401).
- 6.2 Remove the plug from the inspection hatch of the respective stage:
- 6.2.1 For inspection hatches at the 1st stage vane assembly 2 and 3rd stage vane assembly 3 similar to Item 5.3.
- Note If hatches with threaded plugs are used on the case with HPC 1st, 2nd ad 3rd stage vanes, proceed as follows:
 - 1) Unlock the inspection hatch plug.
 - 2) Install the wrench 19-590 assembled with the adapter and torque wrench in the recess of the plug.

- 3) Unscrew the plug, counterclockwise, and carefully remove from the engine.
- 6.2.2 For inspection hatches at the 5...12th stage vane assemblies proceed as follows:
 - unlock the inspection hatch plug;
 - install the wrench 32-27-899 on the plug;
 - unscrew the plug, counterclockwise, and carefully remove from the engine.
- 6.3 Insert the endoscope in the hatch between the guide vanes with the direction of view towards the air intake and inspect the blades from the trailing edge side. Turn the endoscope 180 degrees and inspect the next stage blades from the leading edge side.
 - Notes 1 Inspect the blades similar as described in Items 5.4, 5.5 as relevant for each stage.
 - 2 Acceptable for operation without rectification are the blades with corner bends at the tip part of the blade with the size of 1x1 mm, max:
 - 1st stage blades from the trailing edge side;
 - 2...13th stage blades at the leading and trailing edges.
- 6.4 The acceptable number of blades containing the damages specified in Items 5.4, 6.3 per the impeller:
 - Stages 1, 2-6 blades, max;
 - Stages 3, 4, 5 8 blades, max;
 - Stages 6, 7, 8 10 blades, max;
 - Stages 9...13 12 blades, max;.
- 6.5 Inspect 3...12th stages spacer rings.
 - CAUTION. MARKS OF VANES RUBBING AGAINST THE SPACER RINGS SURFACE AND A VISIBLE GAP BETWEEN THE SPACER RING AND THE IMPELLER DISK ARE NOT ACCEPTABLE.
- 6.6 Carefully retract the endoscope from the hatch.
- 6.7 Install the plug of the respective stage into the inspection hatch:
- 6.7.1 For inspection hatches at the 1st stage vane assembly 2 and 3rd stage vane assembly 3 similar to Item 5.7.

Note – If hatches with threaded plugs are used on the case with HPC 1st, 2nd ad 3rd stage vanes, proceed as follows:

- 1) Install the plug on the inspection hatch of the respective stage and screw it clockwise by hand.
- 2) Tighten the plug with the wrench 19-590, the adapter and the torque wrench. The torque is 2.0...2.5 kgf·m.
- 3) Lock the plug with wire Type 44 (070.00.03, page 6, Figure 17).
- 6.7.2 For inspection hatches at the 5...12th stage vane assemblies proceed as follows:
 - a) install the plug of the respective stage on the inspection hatch and screw it clockwise by hand.
 - b) tighten the plug with the wrench 32-27-899, the adapter and the torque wrench. The torque is 2.0...2.5 kgf·m.
 - c) lock with wire the following plugs:
 - inspection hatch plugs 4, 5, 6 Type 44 (070.00.03, page 6, Figure 17);
 - inspection hatch plugs 7, 10 Type 41 (070.00.03, page 6, Figure 14);
 - inspection hatch plugs 8, 9 Type 42 (070.00.03, page 6, Figure 16).

7 Inspect the 13th stage blades from the trailing edge side:

- 7.1 Unlock, unscrew and remove the bolts 12 (2 pcs.) used to secure the plug 11 of combustor diffuser hatch from the engine left or right side. Remove the plug with a gasket.
- 7.2 Remove the gasket from the plug.
- 7.3 Insert the endoscope in the hatch of combustor diffuser and install between the nozzle vanes with the direction of view towards the air intake. Inspect the 13th stage blades from the trailing edge side.
- 7.4 Carefully retract the endoscope from the hatch.
- 8 Install the plug 11 on the combustor diffuser hatch:
- 8.1 Install a new gasket in the plug 11 recess.
- 8.2 Install the plug 11 on the combustor diffuser hatch and secure it with bolts 12 (2 pcs.) with new locking washers.

- 8.3 Tighten the bolts 12 used to secure the plug 11. The torque is 0.8...1.3 kgf·m.
- 8.4 Lock the bolts with new locking washers Type 22 (070.00.03, page 2, Figure 2).
- 9 Install the removed panels "1-4", "1-5", "2-3", "2-4", "3-3" and "3-4" (072.70.00, page 403)
- 10 Remove the fixture for works in the fan duct (072.70.00, page 209).
- 11Remove the handle or electric drive for HPC rotor turning from the engine accessory drive gearbox (072.00.00, page 607 or 609).
- 11a After completion of works:
 - take away the mats from the inner surface of the front casing and TRU doors;
 - remove the fixture for works in the fan duct (072.70.00, page 209);
 - inspect the fan duct for foreign objects and mechanical damage;
 - unlock the thrust reverser in the "Forward Thrust" position (073.21.00, page 203 or 072.80.00, page 209).
- take away the banner prohibiting engine starting from the cockpit.
- 12 Make a record of the works performed in the engine log book.
- 13 Generate an HPC blades inspection report (see Annex A) in three copies, forward one copy to both the engine design and engine manufacturing organizations.

Works to be performed in case of noncompliance with Technical Requirements

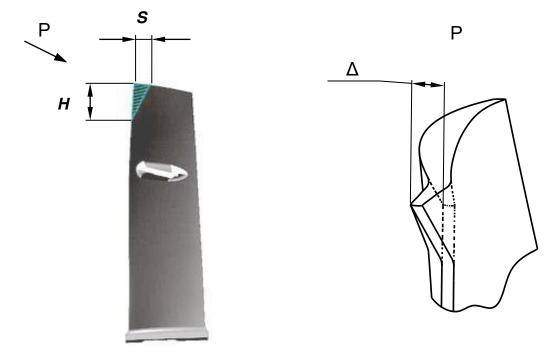
Check

<u> </u>	Instrumentation	Tools and fixtures	Consumables	
	LYMPUS Iplex GX/GT video	Socket wrench head 19-916	Cloth with finished edges	
	ndoscopic system with a changeable probe	S=10 (with internal square 10)	of 0.4 x 0.4 m made of cotton fabric	
	ith a diameter of 6 mm, the length of the orking part is 2 m with a set of	Head for HPC plugs removal and installation 32-27-	GOST 29298-2005	- 2 pcs.
	nangeable lenses.	899 $S_{sq}=10$ (inner)	Nefras – C50/170 GOST 8505-80 or	
	he video endoscopes with similar	Box wrench 19-140 S=10x12	Nefras – C2-80/120	
	erformance may be used if the engine	Torque wrench M=0,31,5 kgf·m		- 0.5 1
	esign organization confirms the	84-27-827-01	Lubricant CIATIM-221 (ЦИАТИМ-221)	0.1.1
p	ossibility of their use (070.00.07).	Torque wrench M=1,05,0 kgf·m 84-27-828	GOST 9433-80	- 0.1 kg
		Wrench for quick-release plugs removal 19-01(M8)	Wire 0,8-TC-1-12X18H10T GOST 18143-72	- 1.5 m
		S_{sq} =10 (inner)	Ethyl alcohol	- 1.5 111
		Cutting pliers Type1X2H8	•	- 0.05 1
I		ТУ39-26-0030578 87514-2009		
		Hammer 19-801.		
		Adapter 93-19-810		
		(outer square 10 x inner square 13)		
		Flat-nose pliers 7814-0266 И Н6.Х.б GOST 5547-93		
		IGV vanes turning device		
•		93-19-811 (19-593) – 2 pcs.		
		Swivel handle 19-8028 $S_{sq.}$ =10		
		Portable light 93-27-844		
		Rod 19-011(extension to the swivel handle 19-8028)		
		Rod 19-8027 $S_{sq}=10$ (inner)		
		Caulking tool 84-19-022 (for plate lock locking)		
		Wrench for plugs removal 19-590 S=8		

	Description of Operations and Technical Requirements	Works to be performed in case of noncompliance with Technical Requirements	Check	
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Table 2 – Values of HPC 1st stage blade airfoil section corner bend at the leading edge side

Blade airfoil section	Bend corner	
Height from the tip, (H), mm	Width from the leading edge (S), mm	
25, max	10, max	10°, max



Check

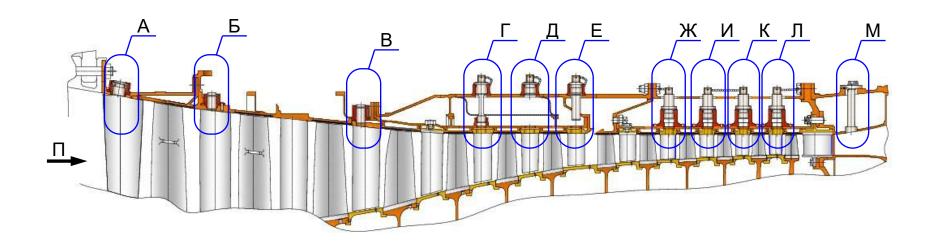
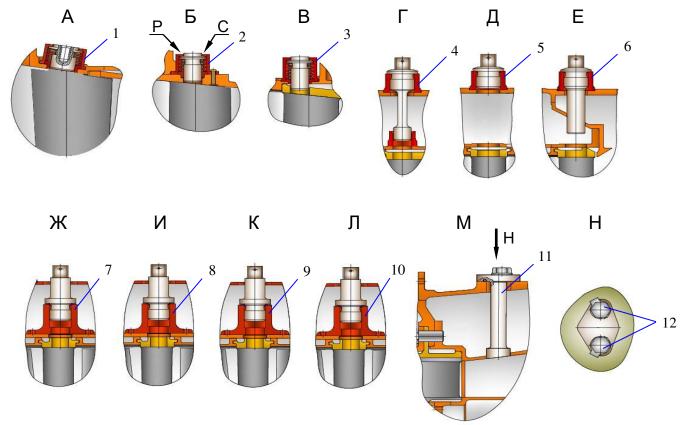


Figure 207 – Diagram of HPC blades and spacer rings inspection hatches location (Part 1 of 3)



- 1 Stage 1 blade inspection hatch, IGV case
- $2-Stage\ 1\ \&\ 2\ blade\ inspection\ hatch-St.1\ vane\ assembly$
- 3 Stage 3 & 4 blade inspection hatch St.3 vane assembly
- 4 Stage 5 & 6 blade inspection hatch St.5 vane assembly
- 5 Stage 6 & 7 blade inspection hatch St.6 vane assembly
- 6 Stage 7 & 8 blade inspection hatch St.7 vane assembly

- $7-Stage \ 9 \ \& \ 10$ blade inspection hatch St.9 vane assembly
- 8 Stage 10& 11 blade inspection hatch St.10 vane assembly
- 9 Stage 11 & 12 blade inspection hatch St.11 vane assembly
- 10 Stage 12 & 13 blade inspection hatch St.12 vane assembly
- 11 Combustor diffuser hatch plug
- 12 bolt

Figure 207 – Diagram of HPC blades and spacer rings inspection hatches location (Part 2 of 3)

	Works to be performed	
Description of Operations and Technical Requirements	in case of noncompliance with	Check
	Technical Requirements	

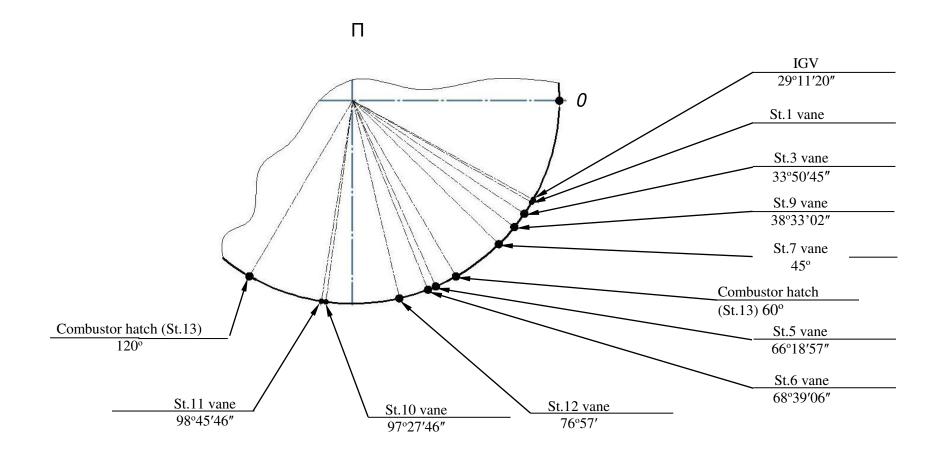


Figure 207 – Diagram of HPC blades and spacer rings inspection hatches location (Part 3 of 3)

Check

Annex A (mandatory) HPC Blades Inspection Report

Product – PS-90A Engine...

Engine No.: ...

TTSN (hours/ cycles): ...

A/C model: ... Powerplant number: ...

Work type (maintenance, post-flight inspection, etc.): ...

Work completion date (DD. MM. Year): ...

Work performed by (Name): ...

Operator representative (Name): ...

Table 4 – HPC Flowpath Inspection Results

		•		Damage description *										
Stage number	Total number of blades containing damage, pcs.	Total number of blades with rectification, pcs.	l blade	Type of damage, (N – nick; D – dent of the airfoil section; B – bend of edge; BC – bend of corner) and	Size from blade tip (l), mm		ge size. <i>L_{max}</i>), m	II lamth at dant			er benc ax, S _{max}		o and large	Сошрнансе
			Con	their quantity on the blade		K _{max}	L_{max}			H_{max}	S_{max}	Δ	Specs	NO

^{* -} Columns should be filled depending on the type of damage (nicks, dents on the airfoil section, edge bending or corner bending), indicating their number and quantity per one blade, and must be accompanied by photos or video records.